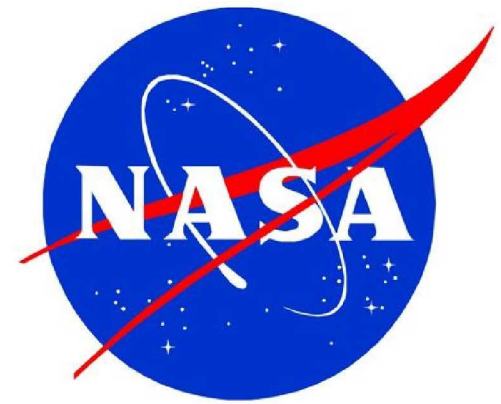


# **DRIVE RIG MUFFLERS FOR MODEL SCALE ENGINE ACOUSTIC TESTING**

Testing of air breathing propulsion systems in the 9x15 foot wind tunnel at NASA Glenn Research Center depends on compressed air turbines for power. The drive rig turbines exhaust directly to the wind tunnel test section, and have been found to produce significant unwanted noise that reduces the quality of the acoustic measurements of the model being tested. In order to mitigate this acoustic contamination, a muffler can be attached downstream of the drive rig turbine. The modern engine designs currently being tested produce much less noise than traditional engines, and consequently a lower noise floor is required of the facility. An acoustic test of a muffler designed to mitigate this extraneous noise is presented, and a noise reduction of 8 dB between 700 Hz and 20 kHz was documented, significantly improving the quality of acoustic measurements in the facility.



# Drive Rig Mufflers for Model Scale Engine Acoustic Testing

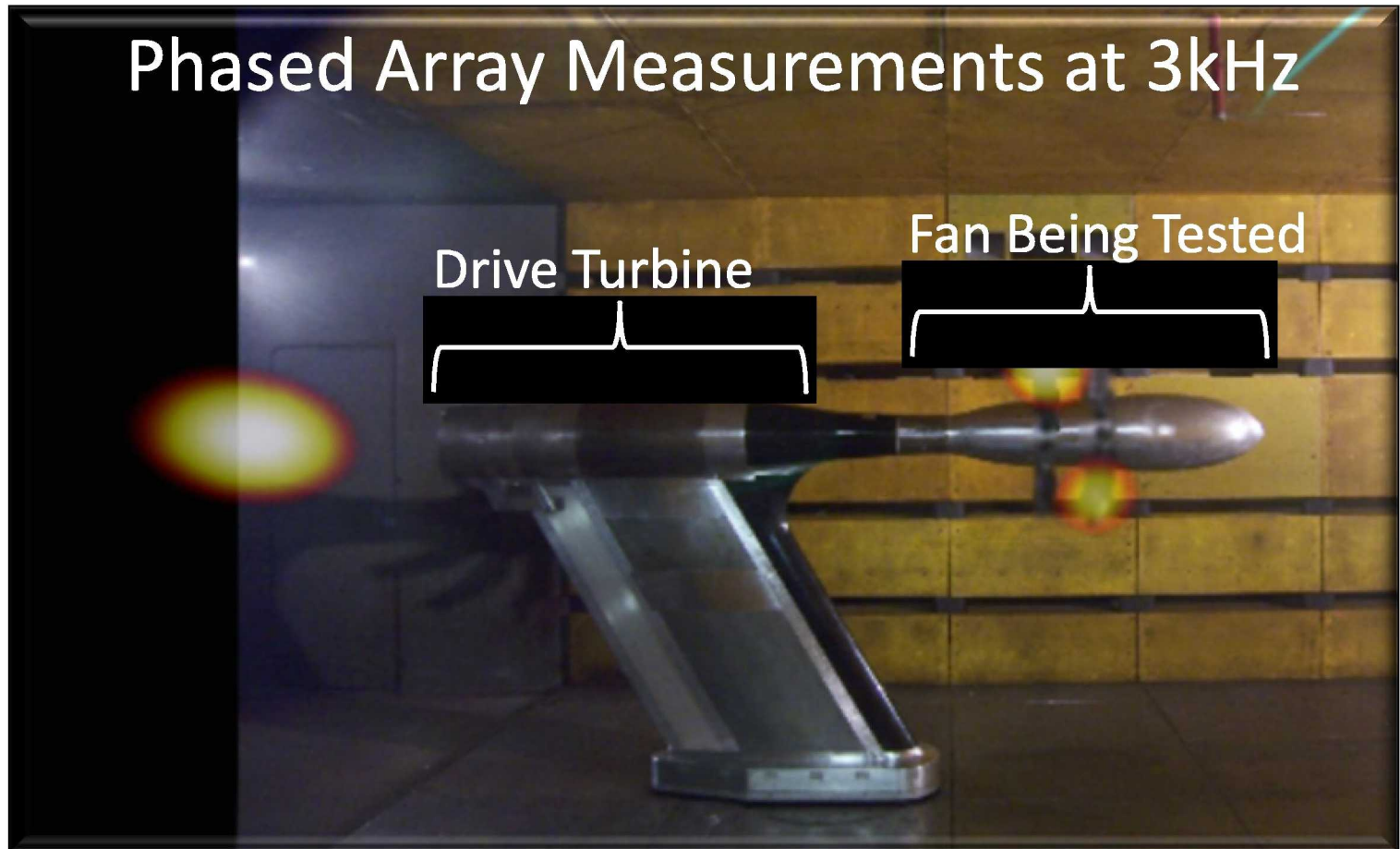
David Stephens

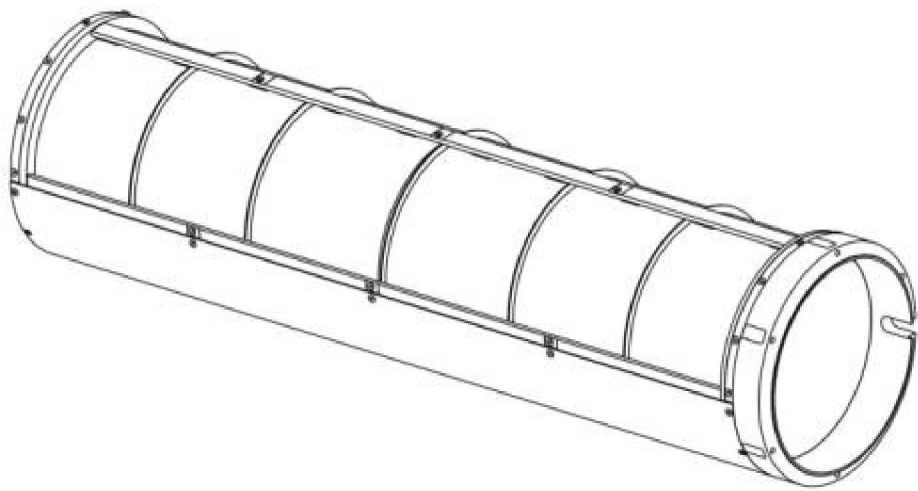
Acoustics Branch

NASA Glenn Research Center

# Compressed Air Turbines used for Model Scale Engine Testing

- ORPR: Two, two stage, 400 HP, 8 inch diameter turbines
- Up to 450 PSI air and up to 20 lbm/s





# Muffler Specifications

Target Performance:

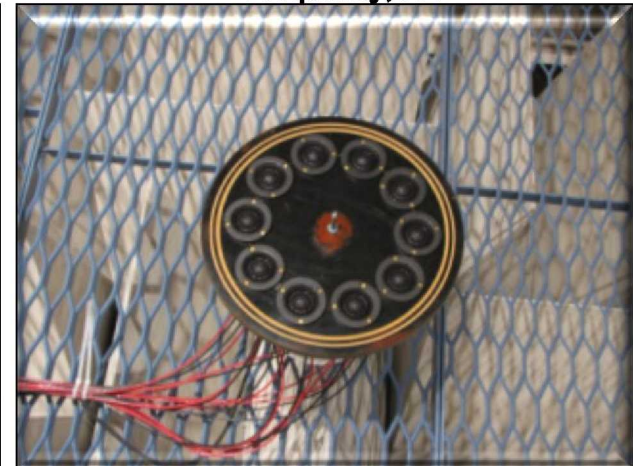
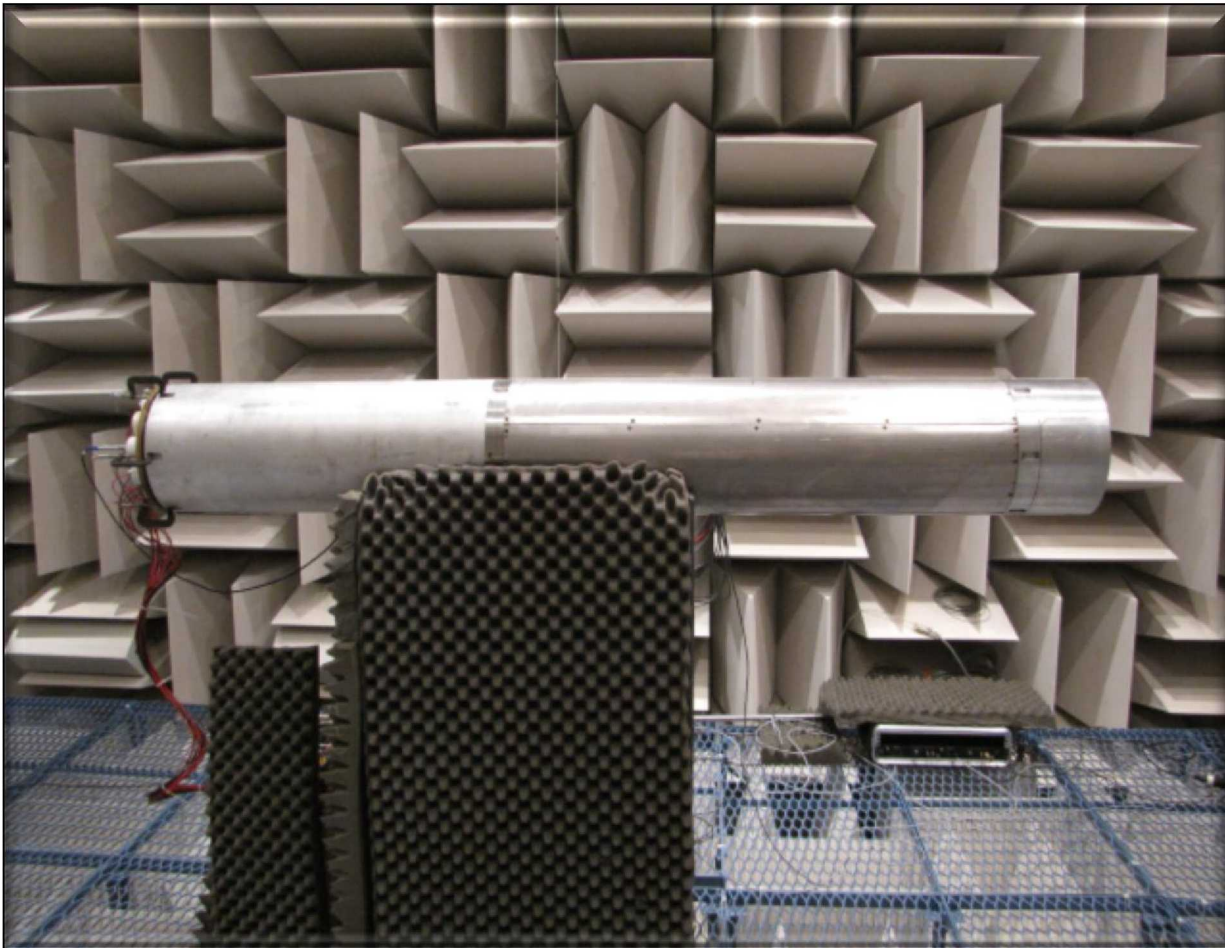
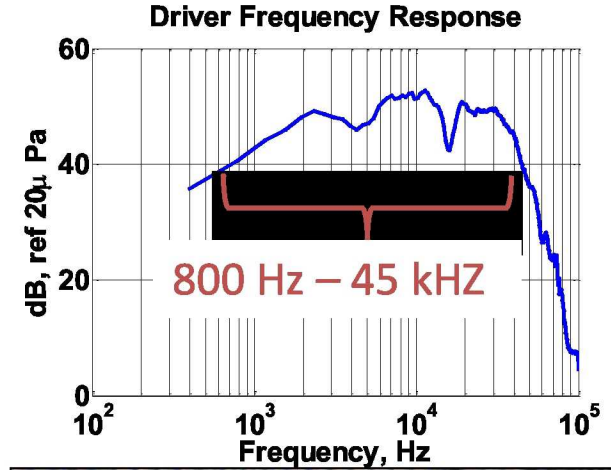
10 dB attenuation 15kHz-25kHz  
(1/5 scale models)

|                   |  |
|-------------------|--|
| Length            | 45 in  |
| Inside Diameter   | 10.5 in  |
| Treated Length    | 42 in  |
| Acoustic Material | Unwoven Nomex<br>.65" thick<br>79 oz/sq-yd                                   |
| Perforate Sheet   | 300 Series Stainless<br>0.036" thick<br>1/16" diameter holes<br>41% porosity |





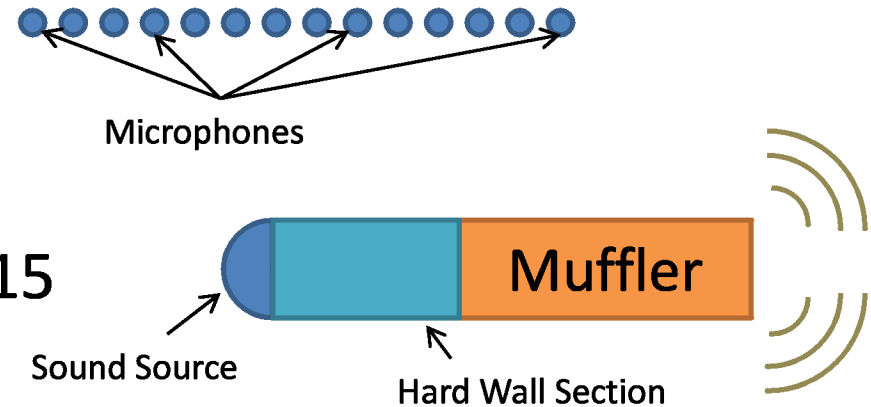
# Acoustic Testing Laboratory (no flow)



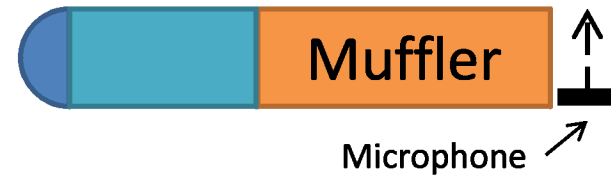
# Microphone Configurations

## Acoustic Treatment vs Hardwall

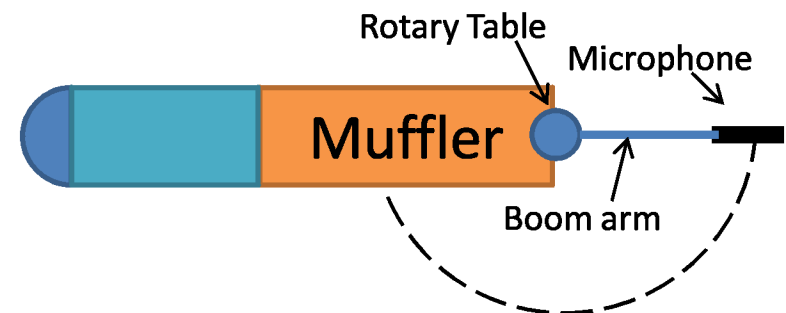
- **Sideline Test**
  - Replicates geometry in 9x15



- **Exit Plane Survey**
  - Documents insertion loss



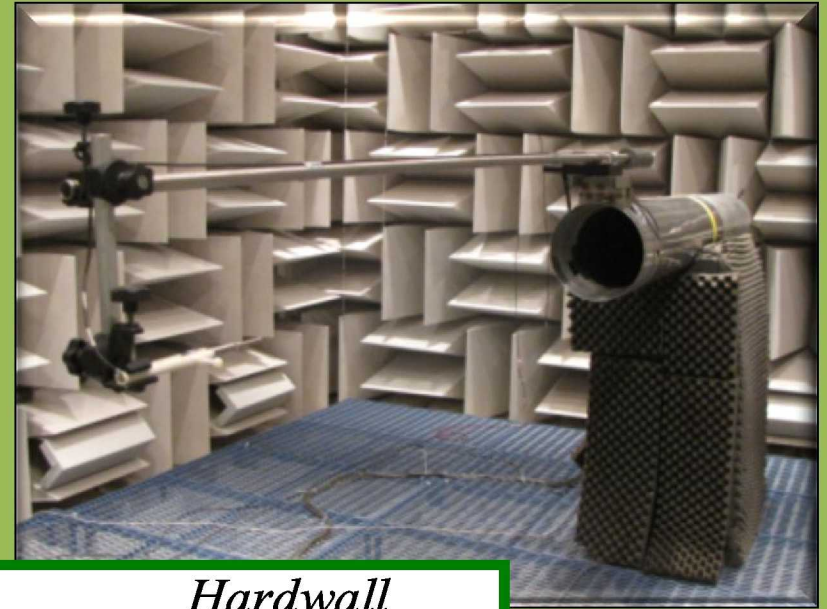
- **Directivity Survey**
  - High spatial resolution



Sideline

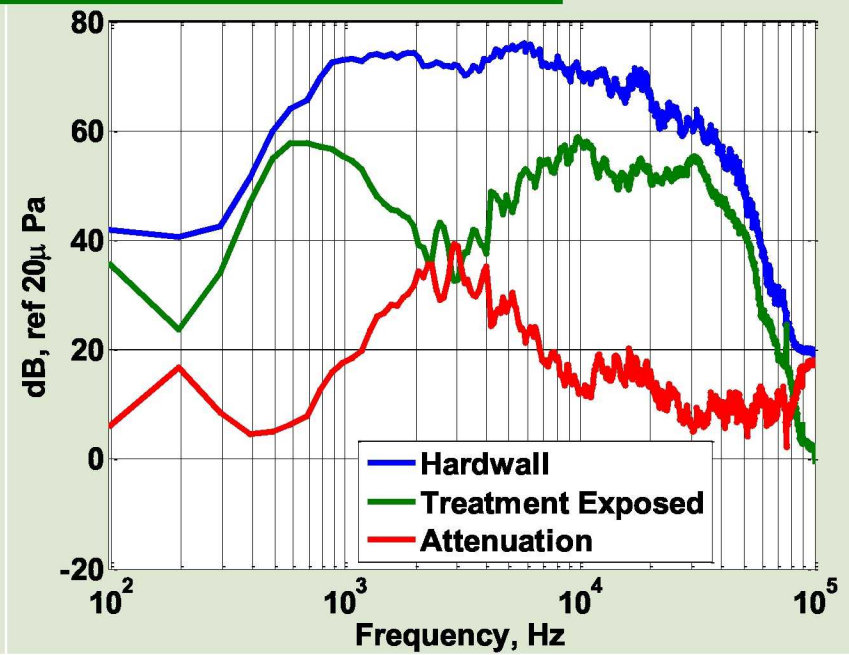
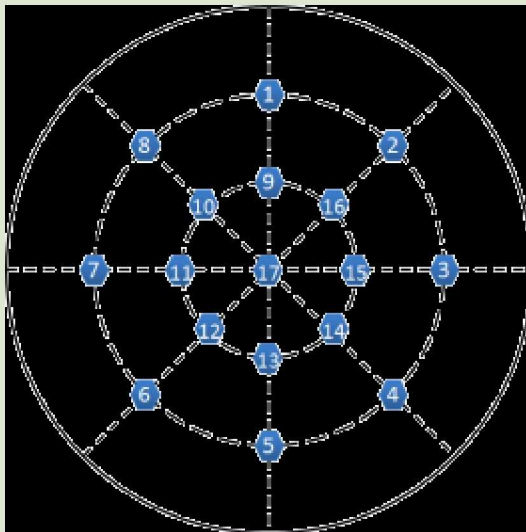


Directivity



$$\text{Attenuation (dB)} = 10 \log_{10} \frac{\text{Hardwall}}{\text{Treatment Exposed}}$$

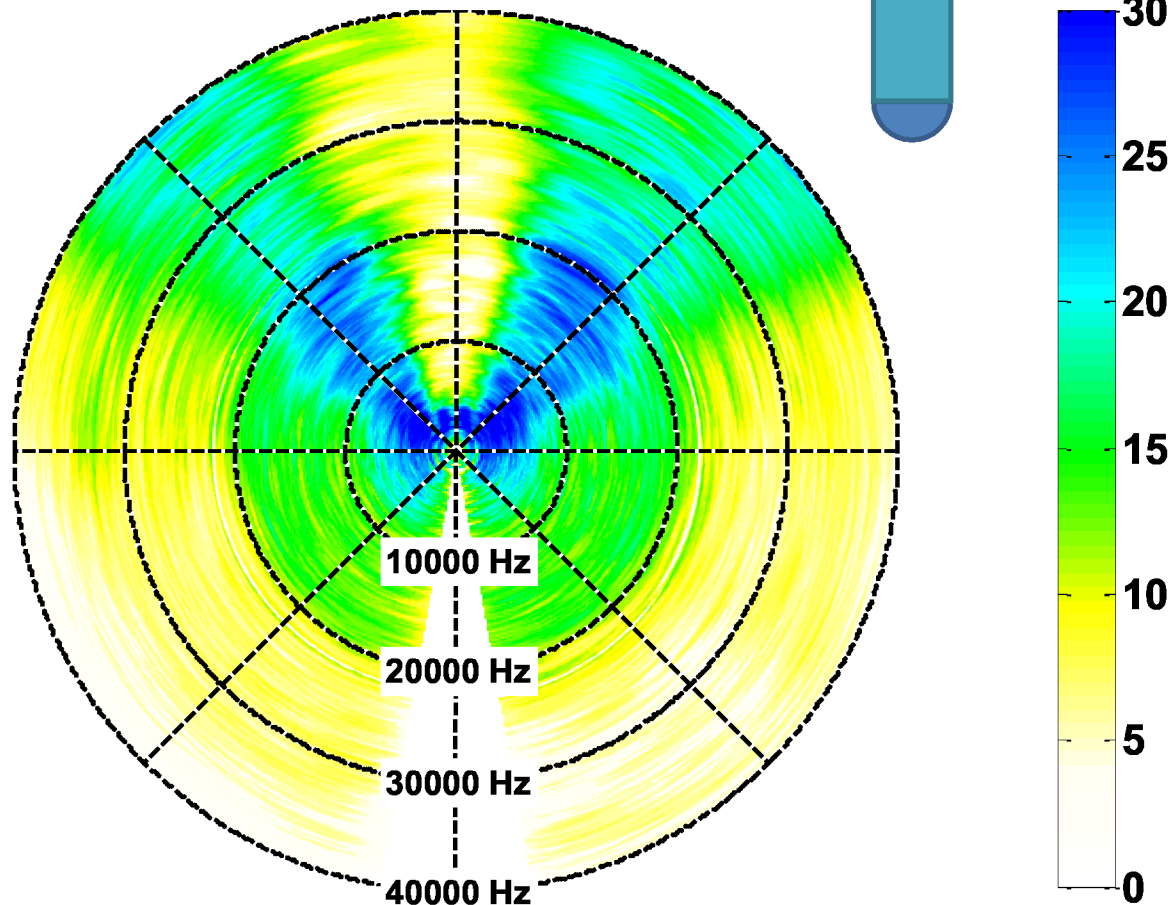
Exit Plane





# Attenuation Directivity

- Peak attenuation of 39 dB at  $\pm 20^\circ$  and 3.2 kHz
- $\sim 25$  dB attenuation at  $\pm 45^\circ$  up to 20kHz
- Around 15 dB attenuation at virtually all angles and frequencies below 20kHz





# Wind Tunnel Measurements

- Nearly 10 dB improvement measured in tunnel
- Higher background noise level
- Performance objectives essentially met

*Study funded by  
NASA Environmentally Responsible Aviation (ERA) Project*

